

Errata

Aerodynamic Design of Axisymmetric Hypersonic Wind-Tunnel Nozzles Using a Least-Squares/Parabolized Navier-Stokes Procedure

John J. Korte
NASA Langley Research Center,
Hampton, Virginia 23665

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THE following figures were too small in the published paper. The AIAA Editorial Staff regrets any inconvenience it has caused our readers.

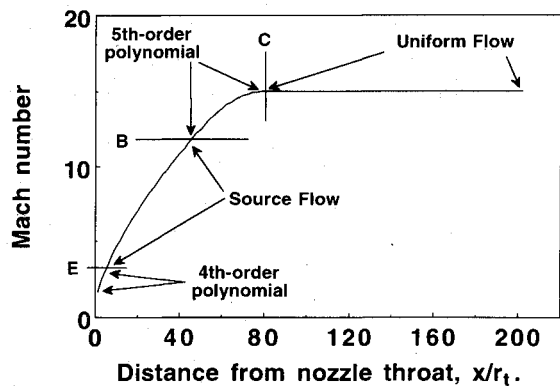


Fig. 1 Design centerline Mach number distribution.

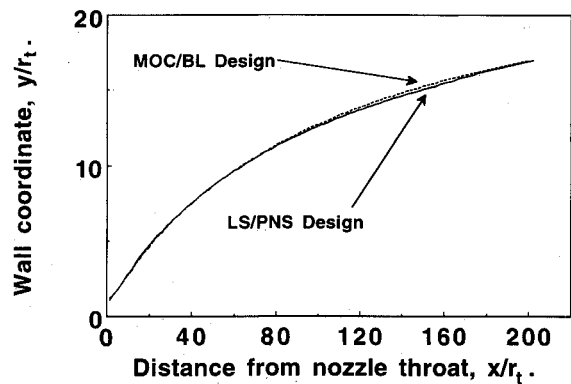


Fig. 4 Comparison of wall contours for Mach 15 design.

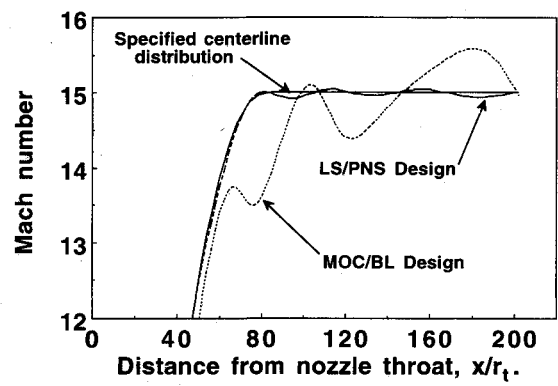


Fig. 5 Centerline Mach number distribution.

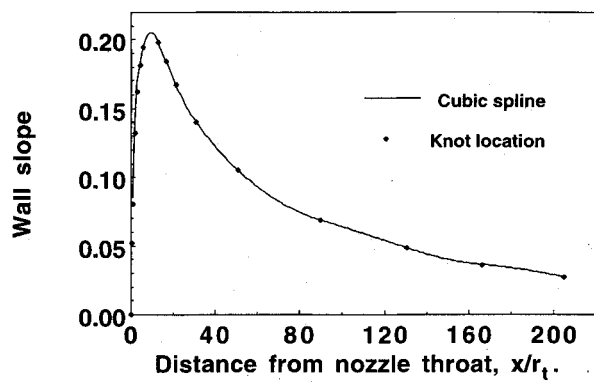


Fig. 2 Wall slope computed using a cubic spline.

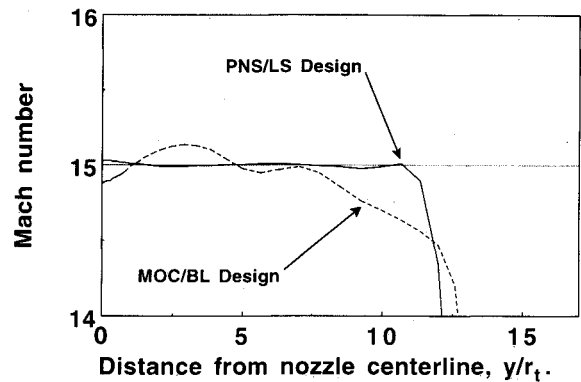


Fig. 6 Nozzle exit Mach number profiles.

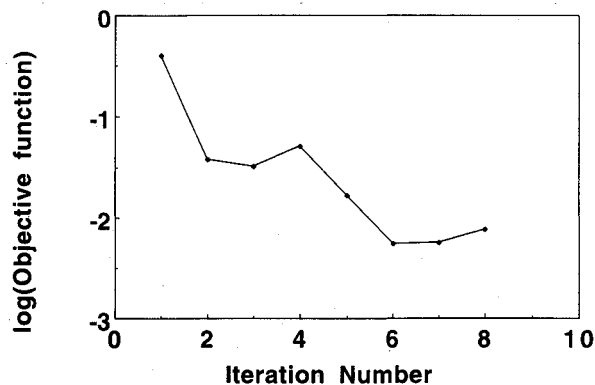


Fig. 3 Reduction of objective function for Mach 15 design.

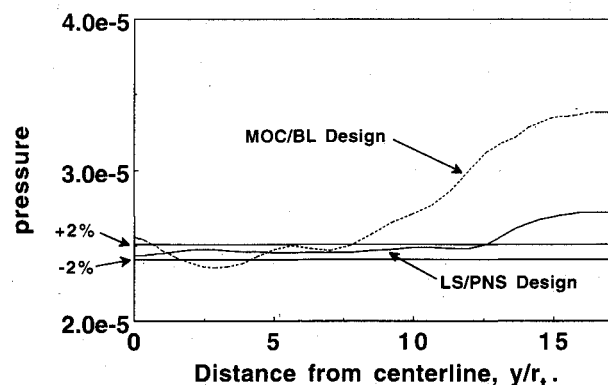


Fig. 7 Nozzle exit pressure profiles.

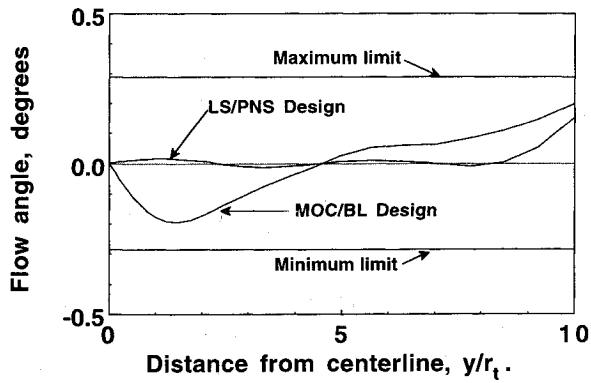


Fig. 8 Flow angle.

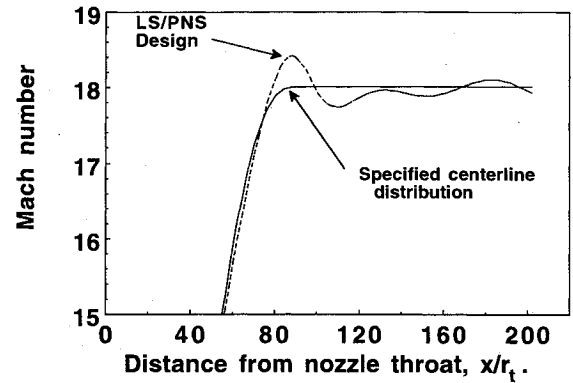
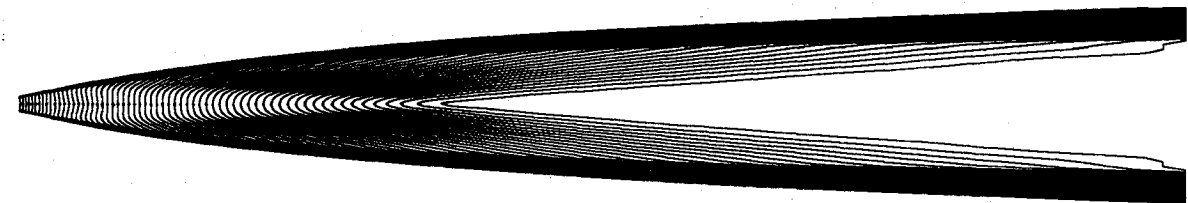
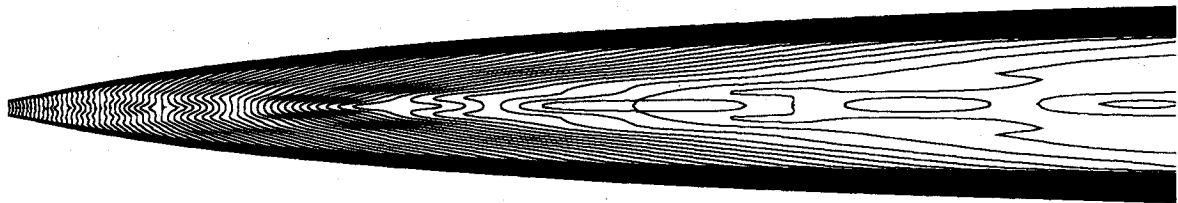


Fig. 12 Comparison of specified and computed centerline Mach number distribution for Mach 18 design.



Least-squares/parabolized Navier-Stokes design



Method-of-characteristics/boundary layer design

Fig. 9 Comparison of Mach number contours for LS/PNS and MOC/BL designed nozzles, 0.2 Mach increment.

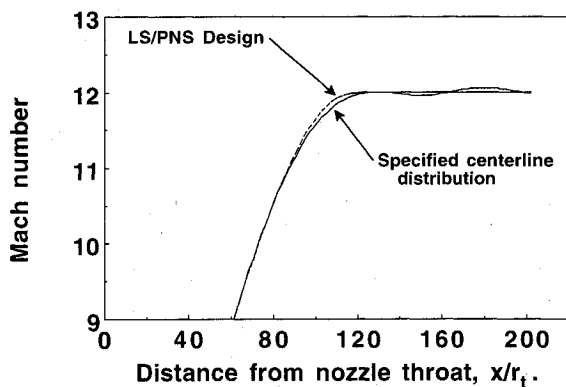


Fig. 10 Comparison of specified and computed centerline Mach number distribution for Mach 12 design.

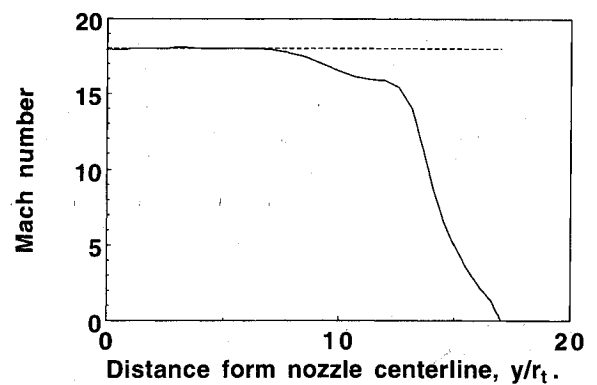


Fig. 13 Comparison of Mach exit profile with design Mach number Mach 18 design.

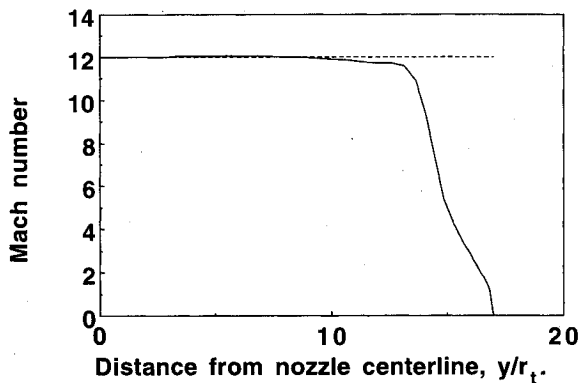


Fig. 11 Comparison of Mach exit profile with design Mach number, Mach 12 design.